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AIR FORCE FLIGHT DYNAMICS LABORATORY

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AIR FORCE SYSTEMS COMMAND

WRIGHT PATTERSON AIR FORCE BASE OHIO



A USER'S MANUAL FOR THE
SEQUENCE ACCOUNTABLE FATIGUE ANALYSIS
COMPUTER PROGRAM

OCTOBER 1973

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FORWARD

This program was prepared by J. M. Potter of the Solid Mechanics Branch and R. A. Noble of the Experimental Branch, Structures Division, Air Force Flight Dynamics Laboratory. This work was conducted in-house under Project 1347, "Structural Testing of Flight Vehicles," Task 134704, "Structural Testing Criteria." This memo covers work accomplished over a time period of 1 October 1972 to 1 May 1973.

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This Technical Memorandum has been reviewed and is approved.

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ABSTRACT

This report presents a detailed description of a computer program to calculate cumulative damage of notched structural members subjected to arbitrary spectra. The Sequence Accountable Fatigue Analysis computer program develops its sequence sensitivity by tracking residual stresses local to a notch throughout the spectrum of loads. Residual stress relaxation analysis is included to increase the generality of the results. An example spectrum and resulting cumulative damage analysis are illustrated.

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SYMBOLS

res	Residual stress
Jmax	Maximum local stress level
σ_{\min}	Minimum local stress level
σys	Yield stress
^σ res _{EQ}	Equilibrium component of the residual stress
ε _t	Local strain, total
ε _e	Elastic component of total strain
ε p	Plastic component of total strain
S max	Maximum applied stress level
S _{min}	Minimum applied stress level
Smean	Mean applied stress level, $(S_{max} + S_{min})/2$
Kt	Elastic stress concentration Factor
D	Damage
ı	Integer describing the level number
NEP	Equilibrium period, number of cycles for the local stresses to return approximately to the equilibrium conditions following an overload
CI.	Residual stress relaxation constant
E1, E2	Relaxation function exponents
N	Number of cycles
E	Modulus of elasticity
N.	Number of cycles of life at a given stress or strain cycling level
E P	Strain intercept at one reversal on a log ep-log life curve
10 11 11 11 11 11 11 11 11 11 11 11 11 1	Slope of the log ep-log life curve
1.7	

SECTION I

INTRODUCTION

Cumulative damage analyses based upon the local stress-strain behavior at a notch appear to be reasonably successful in anticipating trends in fatigue life behavior of notched specimens subjected to spectrum loading (1-6). The type of behavior that usually occurs is that peak tensile loads tend to increase the fatigue life and peak compressive loads tend to decrease the life of notched structures compared to structures experiencing load spectra not having those peaks (5,6). Local behavior analyses, such as those developed by Smith (7) and Neuber (8), help to explain this phenomenon as being a result of the tensile peak load creating a compressive residual stress at the notch and, conversely, the compressive peak creating a tensile residual stress. The change in life occurs because the residual stress state modifies the subsequent damage accumulation rates.

The Sequence Accountable Fatigue Analysis computer program was developed to incorporate the local stress-strain approach with a recent residual stress relaxation analysis (6) in order to improve the sequence sensitivity of cumulative damage analysis. This technical memorandum presents the details of the resultant computer program and an example of its use. The correlation of predictions made with this analysis to actual results of tests experiencing spectrum loading is presented by Potter (9) and Potter and Noble (10).

SECTION II

PROGRAM OUTLINE

The Sequence Accountable Fatigue Analysis traces the stress-strain behavior local to a notch throughout an applied load spectrum and calculates the damage based on the local experience. The computer program is divided generally into the four parts or modules outlined in Fig. 1.

The basic input data for the material, specimen geometry, fatigue behavior qualities and spectrum, are developed in Module I. Module I is discussed further in Section III. Module II takes the input information and determines the local stress-strain behavior. Module III references the Range Pair Counting Method Subroutine to cycle count the local stress spectrum. Module IV determines the damage in the local stress-strain spectrum.

The basic analyses used in Modules II, III and IV are presented below.

Module II - Local Stress-Strain Behavior - The analysis used during the determination of the local stress behavior during the spectrum of loading is a combination of analyses developed by Smith (7), Namber (8) and Potter (6). Smith's simple analysis indicated that the residual stress could be approximated by assuming that the initial atress-strain behavior was elastic upon unloading following plastic flow. Thus, the residual stress could be calculated knowing the

maximum local stress and the maximum applied stress as in Eq. 1 and in Fig. 2.

$$\sigma_{res_i} = \sigma_{max_i} - K_t S_{max_i}$$
 (1)

The Sequence Accountable Fatigue Analysis computer program currently incorporates elastic-perfectly plastic stress-strain behavior. Therefore, σ_{\max_i} is equal to the yield stress. For the cycles immediately following the peak stress, the residual stress determined in Eq. 1 modifies the elastic solution as shown in Eqs. 2 and 3 (provided that the following maximum applied stress is less than S_{\max} and that there is no change in the residual stress due to a minimum applied stress causing reversed yielding).

$$\sigma_{\max_{i}} = \sigma_{\text{res}_{i-1}} + K_{t} S_{\max_{i}}$$
 (2)

$$\sigma_{\max_{i}} = \sigma_{\operatorname{res}_{i-1}} + K_{t}S_{\max_{i}} \qquad (3)$$

The analysis developed by Neuber (8) has been extended to cyclic loading by Wetzel (2) and Wetzel, Morrow and Topper (3) and used by many others (1,4-6) primarily to determine local stress-strain behavior. It is used in this program only to calculate plastic strains occurring when the residual stress undergoes a step change. The plastic strain calculation routine is accessed only when the omaxior or mini terms in Eqs. 2 and 3 exceed tensile or compressive yield stress levels, respectively. Figure 3 illustrates the calculation of the plastic strain.

The local stress-strain behavior, according to Wetzel (2) is related to the applied load by Eq. 4

$$\sigma \cdot \varepsilon = (K_t S_{max})^2 / E$$
 (4)

The plastic strain can be found by subtracting the elastic component from the total strain.

$$\varepsilon_{\rm p} = \varepsilon_{\rm t} - \varepsilon_{\rm e} = (K_{\rm t} S_{\rm max})^2 / E \cdot \sigma_{\rm max} - \sigma_{\rm max} / E$$

Therefore, the plastic strain associated with S_{max_4} is given in Eq. 5.

$$\varepsilon_{p_i} = (K_t S_{max_i})^2 / E \sigma_{ys} - \sigma_{ys} / E$$
(5)

If a residual stress existed prior to this plastic strain excursion, the plastic strain associated with that prior excursion is subtracted from Eq. 5 as shown in Eq. 6.

$$\varepsilon_{p_i} = (K_t S_{max_i})^2 / E \sigma_{ys} - (\sigma_{ys} - \sigma_{res_{i-1}})^2 / E \sigma_{ys}$$
 (6)

A similar calculation is made for plastic strains occurring during the minimum stress peak.

In the analysis developed by Potter (6) the residual stress cyclically relaxes toward zero or an equilibrium residual stress as shown in Fig. 4 according to Eq. 7.

$$\sigma_{\text{res}_{N=1,2,...}} = (\sigma_{\text{res}_{N=1}} - \sigma_{\text{res}_{EQ}}) \exp(N/N_{EP_i} \ln(0.1))$$
 (7)

The N term, the Equilibrium Period, is dependent upon the applied stress and the Residual Stress Relaxation Constant.

$$N_{EP_i} = (C1/(K_t S_{max_i}^{E1} \cdot K_t S_{mean_i}^{E2}))$$
 (8)

The Residual Stress Relaxation Constant, C1, has not yet been experimentally defined but should be a constant for a material.

Module III - Cycle Counting Method

After the local stress and plastic strain behavior is calculated,
the hocal stress spectrum is Range Pair Counted using a computer program
developed by Tischler. (11)

Module IV - Damage Calculation

Damage is calculated separately for the plastic strain excursions and the elastic stress spectrum. The damage is determined from the conventional $D = \sum_{n=1}^{\infty} \frac{1}{N}$ calculation. Damage from each of the plastic strain cycles is determined from the Coffin-Manson (12) form

$$D_i = 1./N_{f_i} = 1./(\epsilon_{p_i}/\epsilon_{f'})^{1/c}$$

Damage from the elastic stress cycles is determined in a similar manner. The maximum and minimum local stress levels are sequentially compared to unnotched S-N data in a Modified Goodman Diagram format. Damage is summed, and failure of the coupon is defined as the event occurring when the summed damage equals unity.

SECTION III

INPUT DATA REQUIREMENTS

In general, each spectrum analyzed will require slightly different programming in order to get the load history into a usable format for the core program. The basic program requires a certain family of information before any analytical predictions can be made. Appendix I contains a program listing for the Sequence Accountable Fatigue Analysis. The subroutine CORE which accesses the subroutines having to do with RPCM, the Range Pair Counting Method, contains the basic analysis. Subroutine SAL reads the data input and then references subroutine CORE. The subroutine SAL shown is one in which a block of cycles is repeated with optional cycles. A list of the input data cards and the resulting analysis is given in Appendix II.

The specific data requirements are given below.

- 1. Stress-Strain Behavior The stress-strain behavior is presumed to be elastic perfectly plastic with the tensile yield stress being equal to the compressive yield stress. The yield stress value used is an average of the monotonic behavior generally being above the 0.2% yield value and below the engineering ultimate strength.
- 2. Residual Stress Relaxation The residual stress relaxation behavior of Eq. 7 and 8 is characterized by Cl, the Residual Stress Relaxation Constant and El and E2, the relaxation equation exponents.

 The Residual Stress Relaxation Constant, Cl, has not yet been adequately determined. It should be a material property if the relaxation function

is correct and must be assumed. A reasonably accurate estimate of the Residual Stress Relaxation Constant for aluminum material falls in the range of 5-20 x 10^6 (cycles) (Ksi) 2 . Further experimentation on the part of the analyst should develop a Cl usable for his set of conditions until actual measurement of residual stress relaxation behavior defines the relaxation function and constants. At present El and E2 are considered to be equal to 1.0.

- 3. Specimen Geometry The elastic K_t value (if available) is entered into the analysis. If that value is not available then an estimate from some other method may be used. In certain cases, a value may be determined from a constant amplitude fatigue test of a similar structure by fitting several values of K_t to the analysis and determining the best correlation as is done with the K_f solution. Once a stress concentration factor, K_t, is determined for a specimen, that value is not changed from test-to-test of the same coupon configuration,
- 4. <u>Load Multiplier</u> Different spectra are presented for analysis in different manners. Some data are presented in percent of maximum stress, others in terms of nominal stress, and others in terms of hending moment. The value of the load multiplier defines the nominal stress history.
- 5. <u>Cumulative Damage Analysis</u> The damage from the range-paired elastic stress spectrum is determined by calculating a simple N value for each level and accumulating the total. The N_{fi} value is determined from unnotched coupon S-N data in the Modified Goodman Diagram format.

The program requires the input of four second order equations describing the maximum and minimum stress levels at lives of 10⁴, 10⁵, 10⁶ and 10⁷ cycles. The coefficients of the equations are derived by least square fitting the S-N data presented in the form of Eq. 9.

$$S_{\text{max}} = A(I)S_{\text{min}}^2 + B(I)S_{\text{min}} + C(I)$$
 (9)

The A, B, and C coefficients for several typical materials are presented in Appendix IV. The S-N data shown was derived from various sources but usually from the MIL-HDBK-5A (13). The C coefficients correspond to the maximum stress level at zero to maximum applied stress conditions on the unnotched coupons.

The damage from the plastic strain cycles is determined using the Coffin-Manson relation to calculate the N $_{\rm fi}$ value. The conventional plastic strain intercept at one reversal and the $\epsilon_{\rm p}$ - life slope values are used in the analysis. Specific measured values from the literature are used when available and typical values when they are not available.

6. Analysis or Test Spectrum - The last information needed is the order and magnitude of application of the spectrum used in the test.

SECTION IV

OUTPUT OPTIONS

The Computer Program prints the following output in the process of the analysis.

- 1. Maximum and minimum applied stress and local stress response through the spectrum. Also printed out is the residual stress, equilibrium stress, applied cycles, and the equilibrium period.
- The elastic local stress history as input into the Range PairSubroutine and the resulting Range Paired spectrum.
- 3. The plastic strain occurrence during the spectrum and the damage associated with each strain reversal.
 - 4. The accumulated damage associated with the plastic strains.
- 5. The Range Paired elastic stress spectrum and the damage associated with each level.
- 6. The accumulated damage associated with the current block of loading including the plastic strain damage and the total damage since the initiation of cycling.

At the option of the analyst, he can print out all the above items or only two. The IPRINT value controls what data is printed.

If IPRINT = 1, all six items are printed for each flight or block.

If IPRINT = 2, all items except 2. above are printed. /

If IPRINT = 3, only items 4. and 6. above are printed.

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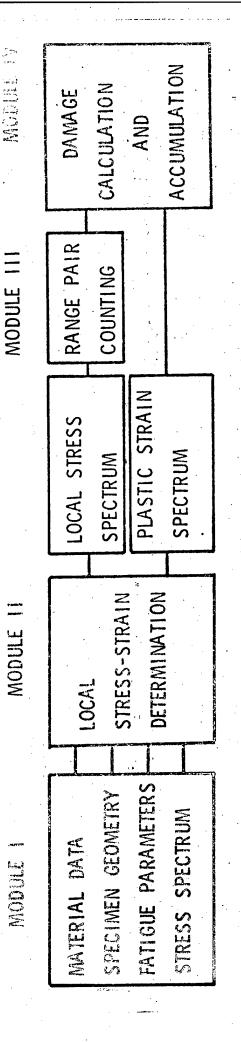
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PROCEDURE USED IN THE SEQUENCE ACCOUNTABLE FATIGUE ANALYSIS FIGURE 1.

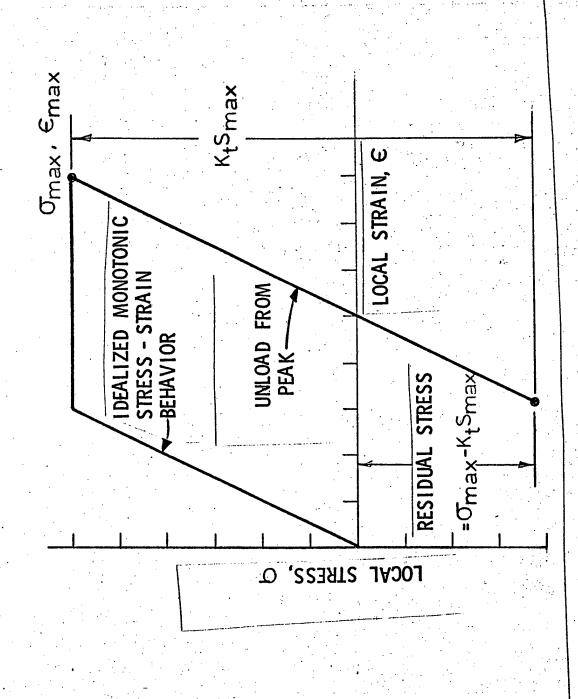


FIGURE 2. METHOD OF DETERMINING THE RESIDUAL STRESS FOLLOWING A PEAK LOAD

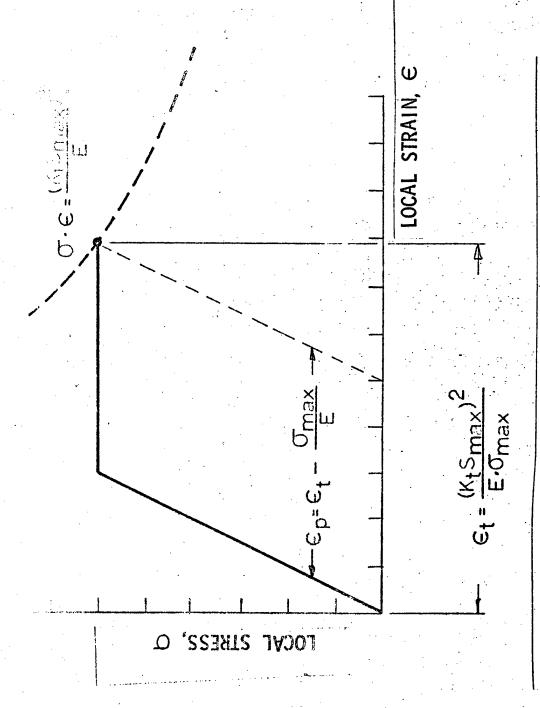
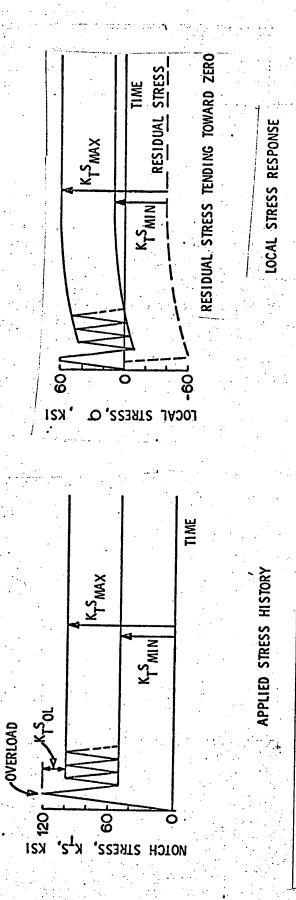


FIGURE 3. METHOD OF DETERMINING PLASTIC STRAIN LEVELS



STRESS RESPONSE FOR APPLIED CONSTANT AMPLITUDE LOADING WITH RESIDUAL STRESS RELAXATION FIGURE 4. LOCAL

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The state of the s	4 F CRMAT(16H1SPECTPU) FROM , 848)
115	
	171153c
	READ(5,10) TH1, TY2, TY3, FFSD, COFMAN, EL MON 10 FORMAT (2A8,3F1A,5, F10,2)
120	WRITE(6,12) TM1, TM2, TYS, EPSD, COFMAN, ELMOD
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	WAITE (6-15)
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4.30	15(6,18)
700	
	20 FORMAT (5X,5H LIFE, 10X,FH 4(11,14X,5H B(T),14X,5H G(I)) WRITE (5,22) (N,4(N),3(N),0(N),N=4,7)
135	22 FORMATICSH 10**, 12, 3518, 5) WRITE (5, 55) TITLE 4, 17 IT F2
1.	SO FORMAT (39HOUMOTOTH) COUPON 3-N DATA DERIVED FRONZSH INFORMATION
18	SEAD 15,14
140	WAIT (6,24)
0.1	HRITE(6, 26)
•	FORMAT (742H WRITE(6,28) C1,E1,E2
145	ורי הי
150	INPUT UP DATA PECULIAR TO A SEQUENCE
155	PEAU(5,32) NBLOCK, JLEVEL,NTYPE 32 FORMAT(3110)
	HAITE(6,34) NBLOCK,JLEVEL 34 FCRMAI(// 110,23H TIMES THROUGH BLOCK OF,110,6H LOADS)
160	33 FORMATIC 5x.13H 10AD 1 THIT = E18 EN
	READIS,36) (ISTEP(K), ITYPETK
165	8 FORMATIC/11H SIEP TYPE-10X,6H SIMIN

1,000,71		01/1/0 TILL 01/0 0000 000	100.000	1 20 1	.
	WRITE(6, 36) (ISTEP(K), ITYPE(K), RTMIN(K), RTMAX(K), RNN(K), K=1, JLEVEL)	RTMAX(K), RNN(K), K=1, JLEVEL)			
	WRITE(6,39) 39 FORMAT(//47H 3LOCK TYPE TYPE TYPE TYPE TYPE	TYPE TYPE TYPE TYPE/)			
170	00 42 JJ=1,N9L06K READ(5,40) IBL06K(JJ), (NN(JJ,KK),KK=1,NT	TYPE	indicates direct provides the second control of the second control		
4	40 FORMAT (915)	- Cheta			
+	42 CONTINUE SUMENNED.	es es resultante de la companya del la companya de la companya del la companya de			
175	SURVO=0.				
	WPITE(6,8)11,12,13,74,15,76,17,18				
- 180	51 FORMAT (774 ART = , (F6.2)) MRITE(6.55)21				
	35 FORMAT(//24H RELAXATION CONSTANT 61= F15.2) IF(18PCM, 6E.2) GO TO 59				
	UBJECTED TO THE	RANGE-PAIR COUNTING TECHNIQUE)			
185	*) CONTINUE				
And the state of t	00 1002 KFL=1,NBLOCK JJJ=1				
190	00 60 J=1,JLEVEL DO 70 KK=1,NTYPE				
And the state of t	TE(INV(CFL,KK), EQ.0) GO TO 60 IF(ITY2E(J),EO,NN(CFL,KK))GO TO 150				:
19		5			
195					
	-3JJ=JJJ+I				
200	NEVEL BORE (KFL)				
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50 60	2 338=0.	
	IF(JA+JB) 270,250,240 CEMOU*IYS)-BBB*BBB/(ELMOO*IYS) CORE	
	RES(1)=1YS-4SMAX 60 TO 290	
65	CORE	
	0.260.270 CORE	
	RES(I)=RES(I=1) 60 T0 290	
70 27	(II) = - ASMEAN	
	SIGHIN (IRAIN) = PES(I) + ASHIN CORE	
	RNCYC(IRAIN) = ENN(J) IF (ASMAX, LE, TYS) GO TO 410	
75	EORES=4SAAX+TVS CORE 7	
410	IF (ASMIN'GE-TYS) GO TO 430 CORE	
80 620	0 4th (1 minutes)	***************************************
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7.7 +	GO TO 446 CORE ABHEABHAX	• • • •
94+	ENEP=C1/(ABM**E1*ABMEAN**E2)	
	WRITE(6, 350) STHAX (J), SIGMAX (IRAIN), SIGMIN (IRAIN), CORE 97	
052	FORMIT(6 (F7,2,1X), F6,2,1X,315,8,5X,16)	
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180	TO 552 STRESSES AND PLASTIC STRAINS W/RESULTING FATTGU 10x,14HPLASTIC STRAIN,10x,10HMAX OR MIN,15X,6HD FROM PLASTIC STRAIN CYCLES L 2,531,533 TRA(JKL) 1/FPSD)**COFMAN		179 1179 1181 1181 1184 1185 1185 1189 1189 1189		
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	X;3HMIN,15X;E14.6)		200		
	WRITE'16, 219) JKL, PLSTRA(JKL), DAM FORMAT (10x,14,12x, F10,5,15x,3H4AX,15x,E14,6)		202 203		
531 CONTINUE WRITE(5,541)		CORE	205 205		
205 541 FORMAT(59X,29H DAYAGE IF(IPRINT,6E,3) GO TO	FROM PLASTIG STRAINS=,E15.8)		206		
WRITE(6,13) 13 FOPMAT(/16X,15H	SIGMAX SIGMIN, 18X, 6H PNCYC, 20X, 25H CYCLES	,	208 209		
210 * ENYCYC)			210 211		
G CALCULATE E	AGE FROM LEAST SOUARE FITTED S-N DATA		212 213		
0.4	MAN DIAGRAM FORMATE		214 215		
215	I-KPHKK		216 217		
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Z20 CYCLES=10.**4	.1.6*TYS) GO TO 310 .**4.	CORE	220 221		

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CORE 222		CORE 225					CORE 232			CORF 235			CORF 240				CORE 245	-		CORE 249				CORE. 255 Core. 256	:	. !	-	CORE 261	CORE 263 CORE 264	CORF 265					
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50 FO 349		50 TO 340	330 M=1++	11 12 12 12 12 12 12 12 12 12 12 12 12 1	CONTINUE	IF (R(7)*R(4)) 339,338,334	4 3F7 # 195 (R (7))	1814847.15.18147 IFIA847.15.1884) 63 TO 335	EXPO=4.+R(4) / (P(4)-R(5))	60 10 359 FYDA 10 (7) 7(B(K) - 2(7))	60 TO 339	2(+)+3(2)+3(9)+8(2)	0.05577857	** (1) c++++19	2(5)+7. + 2(7)	**2+6, *2 (6) **2+7	DELL#4.**SUAK**SUAK4+4.**SU4K**ZUAK**Z DELK#SUAK*SUAK2**SUAK3.*SUAK4*Z	55++3	D01=22, *SUMR2*SUMR4-22, *SUM91**2	DOS=SUM22*SUM23*SUM2N=JUM3*SUM24*SUM3H DO3=SUM2*SUMP3*SUMRPH=SUM32P4*SUMRPH=	EXPOS (001+002+003)/(05L1+05L2+05L3)	7** U EVA 1000 1 TO 100 100 100 100 100 100 100 100 100 10	YOU	_	IF(IP2]NT, GE, 3) 50 TO 600	16X,2(F7.2,1X),16X,F5.0,17X,	CONTINUE	WRITE(6,593)SUMDEL FORMAT(769x,21H DAMAGE PER THIS SET=,515.3	ENN/3YC =, 615.8)	The second secon	eren ere eren eren eren eren eren eren				
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0		RESIDUE SPECTRUM	2*NPKS	CORE	282		
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20 5	-	S OF THE	NPKS + KK	CORE	285	AND THE RESERVE AND THE PERSON AND T	
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25		. 9 AND VALUES OF NSTEP		CORE	290		
25	<u> </u>	AT SIGMAX(•	CORE	291		ar air thin
	8	ICINIUST = II-CINIUST		CORE	293		
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30	COMMON/ASAL/RNCYC (200) + KPM	SYC (200) + KPMAY IPPINT		C02E	295 296		
	CT (CCC) dateNCCCHONNORM		***************************************	CORE	297		
	COMMON/MDECR/RES(1400)	X(1400), IND1, IND2	,IND3,IND4,KIND	CORE	298		·
OF THE STREET, AND ADDRESS OF THE STREET, STRE	COMMON/YCYG/CYCLE (900,2) +7	CLE (900,42), RNECYC (900), NNSTEP (900)		3000	299		
35	CURMUNY ACCUENT OF TAINE	- (90) - 111 F (8)		CORE	301		
	0 = HCNOdN 6666			CORE	. 302		
	00 8000 I = 1,1	1.NPKS		CORE	304		
04	IF (IPRINT	.GE.2)50 T7 103		CORE	305		
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	MRITE(5,25)	PSIS,	, I = 1, NPKS)	CORE	312		
	25 FOFMAT (29X,1	5,10x, £13.6,10X,£13.6,10X,F10.5)		CORE CORE	313		
2 05	SORT THROITER	1040 SPECIFIEM - PULL OUT THOSE PE	AKS AND VALLEYS	CORE	315		
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	TINITING	-		376	
	;		CORE	377	
	S RANGE PAIR COUNT THE ADJUSTED LOAD SPECTRUM		CORE	379 380	
	6000 T = 1 X8 = 0		CORE	381	
] Z		CORE	383	
120			CORE	385 386	
	(3NCYG(I) .GT.		CORE	387	
125	E. 01 GO TO 5 AX(I)		CORE	389 390	
The state of the s	" 70		CORE	301	
	+		CORE	303	
130	KB = 1 G0 T0 1		CORE	395 396	
	5 X3 = SIGHAX(I) X4 = SIGHIX(I)		CORE	398	
135	50		CORE	399	
	0 0		CORE CORE	401 402	
27	K31 = 0 IF (RNCYC(I) .Eq. 1.0) GO TO 6		CORE	403 403	
140	KEY = 1 KIND = 1		CORE CORE	405	
	60 73 415 6 KEY = 0		CORE	408	
145	CALL DESIDE(X1, X2, X3, X4, KEY, I, CYSNO, KCYGEN)		CORE	409 410	
	1000 GO TO (10,10,30),KCYGEN 10 K9 = 1		CORE	411 412	
			CORE	413	
150	F (T .LE		CORE	415 416	
	λ. 		CORE	418	
155	in &		CORE	419 420	
	TO IF (CHIN .NE. 19 GO TO 35		CORE	42 1 422	
			CORF	424	
160	ES(LR+1) = ES(LR+2) =		CORE	425 -426	
	3) = X 2+1) =		CORE	427 428	
165	INDEX(LR+2) = IND2 INDEX(LR+3) = IND3		CORE	429 430	

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	2030 IMPX = 1	10 5700	1000	0 - 0 2 - 6	מפעע	
	IF (LRMAX .LT. 4) GO TO		CORE	480		
926	THE DATE COMME OF PERSONS	•	CORE	489		
			CORE	490		
	CALL DEDRESTLANTANOYNO)		CORE	492		
. 230	5000 IF (LRMAX -LE- 11 GO TO 3000		CORE	767		
	CITY JONEO FILLIANDE DECEMBER CONTRACTOR	- 1		4.95		
	ADDITIONAL GYCLES	WILL VIELD N		498 498		•
235			CORE	499		
	RESMAX = RES(1) RESMIN = RES(1)		CORE	501		
•			CORE	503		
240	00 500 I = 2,LRM4X IF (25S(I) .11. RESMAX) GO TO 490		CORE	505		
A single community and the same of the sam			CO SE	507	Amplitude of the state of the s	
245	500		CORE	509.		
	RESMIN = RES(I). IMIN = I		CORE	511		
	_		CORE	513		
250	1.2		CORE	• • •		
	IF (J. LE. 0) 60 TO 550 CALL CYCRES(RES(J), RES(J+1), 1.0, INDEX(J))		CORE CORE	-		
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	IF (J .ST. LRMAX/ GO TO 575 CALL GYORES(RES(J-1), PES(J),1.0,INDEX(J-1))		CORE	523		
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	RT THE ANALYSIS SPECTRUM TO PRODUSE THE RANGE PAIR	COUNTED SPECT	1 .	531		
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The control of the	10 TO 11 GO 10 4183	CORE
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	IF (J. 680) LKMBX3 60 IO 30U	CORE	787		:
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INDEX (K+1) = INDE	CORE	734
LRMAX = K + 1	CURE	735
	CORE	736
	CORE	737
RES(X) = XI RES(X+1) = X2	CORE	738
"	CORE	740
I NUEX (K) =	CORE	741
-:	3,000	212
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RETURN	CORE	745
+	CORE	746
(U RES(K) = X2	CORE	747
1	COXE	242
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SONT = (TYX) XEVIL	נטענ	751
(K+2) =	1,100 CO3E	753
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APPENDIX III

LIST OF COMPUTER PROGRAM SYMBOLS AND DEFINITIONS

A Coefficient of the x^2 term in the equation of a line on a constant life fatigue diagram where minimum stress is x and maximum stress is y. $(R = Ax^2 + Bx + C - y)$

AA An assigned value of +1. or -1.

AAA A stress used in the calculation of plastic strain.

ABDIF The absolute value of DIF.

ABM The absolute value of ASMAX or of ASMIN, as assigned.

ABMAX The absolute value of ASMAX.

ABMEAN The absolute value of ASMEAN.

ABMIN The absolute value of ASMIN.

ABR4 The absolute value of R(4).

ABR7 The absolute value of R(7).

ABS The name of a routine calling for the absolute value of a quantity.

AKT Stress concentration factor, K

ASE Array of values of ENN, for the plotting subroutine.

ASMAX The product (AKT) (STMAX)

ASMEAN The quantity (ASMAX + ASMIN)/2

ASMIN The product (AKT) (STMIN)

Array of values of ASMIN, for the plotting subroutine.

ASK Array of values of ASMAK, for the plotting subroutine.

AVSCIN Average value of SIGMIN over an interval.

AVSCMX Average value of SIGMAX over an interval.

B Coefficient of the x term. (See A.)

BBB A stress used in the calculation of plastic strain.

C The constant. (See A.)

COFMAN Inverse of the Coffin-Manson slope.

CYCINT The number of cycles in an interval.

CYCLES The calculated number of cycles expected to be indicated on a constant life fatigue diagram for the applied combination of maximum and minimum stress.

C1 The Residual Stress Relaxation Constant (See ENEP.)

DAM Damage.

DECK Decimal or real value of integer K after conversion.

DEL2 A portion of a least-squares-method solution.

DIF The difference between residual stress and equilibrium residual stress. (RES(I) - EQRES)

DO2 A portion of a least-squares-method solution.

DUMMY A variable used in the calculation of the number of cycles to be considered as an interval for relaxation determination.

ELMOD The elastic modulus.

EN The number of cycles from the beginning of the relaxation process to the end of the current interval.

ENEP The number of cycles required for overload residual stress

effect to return to within one-tenth of its original difference from equilibrium conditions.

 $(N_{ep} = C1/(ABM)^{E1}(ABMEAN)^{E2})$

The number of applied cycles at a load level.

ENNCYC The ratio of the number of applied cycles to the number

of cycles to failure. (ENN/CYCLES)

EPSD ICF strain intercept.

EQRES Equilibrium residual stress.

An exponential function depicting the relaxation of residual stress.

The name of a routine calling for the exponential value of a quantity.

EXPO An exponent. The power of 10 which indicates the number of cycles to failure.

E1 Residual stress Relaxation Exponents.

The name of a routine calling for integer-to-real conversion.

I A variable subscript.

IBLOCK The identifying number of a block the blocks being numbered consecutively from 1 to NBLOCK.

IFIX The name of a routine calling for real-to-integer conversion.

IN The number of steps input to the range pair counting subroutine.

IPRINT Value controlling the WRITE statements.

RAIN A counter.

Value controlling entry into the range pair counting subroutine.

ISTEP The identifying step number, the steps being numbered from 1 to NLEVEL.

ITYPE The identifying type number, the types being numbered from 1 to NTYPE.

J A variable subscript.

JA Value of +1 or 0, as assigned for branch determination.

JB Value of -1 or 0, as assigned for branch determination.

JJ An index variable.

JJJ An index variable.

JKL An index variable.

K An index variable.

KK An index variable.

KPMAX The number of steps output from the range pair counting subroutine.

L An index variable.

LMN An index variable.

M An index variable.

N An index variable with values of N=4-7 indicating the power of 10, and thus identifying a particular life cycle curve.

NBLOCK The total number of times to execute a block of loads.

NDECK The number of data decks to be run sequentially,

NFLAG - An integer used as a counter.

NFLAG2 An integer used as a counter.

MLEVEL The total number of steps, or levels, of loads in a block.

NN A subscripted variable used to indicate which types of

loads are experienced in which blocks.

NTYPE The total number of different types.

PLSTRA Plastic strain.

R Residue term in damage calculation.

RES Residual stress.

RNCYC The number of cycles for a level after exitting the range

pair counting subroutine.

SIGMAX Maximum stress.

SIGMIN Minimum stress.

STMAX Maximum applied stress.

STMIN Minimum applied stress.

SUMDEL Summation of damage for a flight.

SUMENN Accumulated total of applied cycles. (Summation of ENN).

SUMNC Accumulated cycle ratio. (Summation of ENN/CYCLES).

SUMR Summation of R(N), N=4,7.

SUMRN Summation of nR(N), N=4,7. n=4.

SUMR2 Summation of $R(N)^2$, N=4,7.

SUMR2N Summation of nR(N)², N=4,7. n=4.

SUMR3 Summation of R(N)³, N=4,7.

SUMR4 Summation of R(N)4, N=4,7.

TITLE1, TITLE2 Identification of the source of the SN data.

TLL Tensile load limit.

TM1, TM2 Material type.

TTYS One-fifth of tensile yield stress.

TYS Tensile yield stress.

T1,T2,T3,T4,T5,T6,T7,T8 Test identifying information.

X Variable equivalent to SIGMIN.

Y Variable equivalent to SIGMAX.

APPENDIX IV

FATIGUE LIFE INPUT DATA FLR SEVERAL MATERIALS

•	• •							•
	YIELD	STRAIN	INVERSE	LIFE,	S-N LI	FE COEFI	CIENTS	
MATERIAL	STRESS	INTERCEPT	OF SLOPE	10 ¹	A(I)	B(I)	C(I)	
	•					. • • •		
2024-T4	58.	0.4	-1.836	4	0020	.2091	62.6	
	• •			5	0032	.4366	51.4	
	•			6	0035	.6207	42.2	
				7	0042	.7003	36.1	
						0001	FF 0	
2219-T851	55.	0.4	-1.836	4	0022	.2204	55.8	
				5 .	0018	.3320	48.3	
• •				. 6	0015	.4628	39.7	٠,
		• •		7	0024	.6420	31.7	٠.
7075 -T 6	72.	0.4	-1.836	4	0020	.2801	71.7	
1012-110	12.	0.4	-1.000	5	0022	.5154	56.3	
		•		6	0014	.6141	44.6	••
			•	7	0013	.6838	38.1	
•		•	•		0013	•0000	30.1	
RQC-100	125.	0.54	-1.493	4	0.0	.2136	98.3	;
				5	0.0	.2927	88.5	
		•		6	0.0	. 3669	79.1	
	•		•	7	0.0	.4376	70.3	٠.;
							.	:
Man-Ten	55.	1.11	-1.667	4	0.0	.2257	63.5	. :
	:	•	*	5	0.0	.3520	53.1	
*				6 7	0.0	.4669	43.7	
					0.0	.5678	35.4	;
4340 Steel	160.	0.4	-1.836	4	0002	.2567	162.4	
				5	0007	.5248	126.9	٠.
				6	0005	.5557	113.5	. '
				7	0005	.5557	108.5	
						•	•	
T1-6-4	158.	0.4	-1.836	. 4	0009	.2368	154.2	
				· 5	0006	. 4640	110.3	
				6	0000	.4650	88.9	:
				7	.0001	.4752	84.2	٠.